



A DOCPHOENIX

INCOMING

ACPA	Continuing Prosecution Application
AP.B	Appeal Brief
C680	Request for Corrected Notice/Allowance
C.AD	Change of Address
CFILE	Request for Corrected Filing Receipt
COCIN	Papers filed re Certificate of Corrections
CRFD	Computer Readable Form Defective
CRFE	Computer Readable Form 'ENTERED'
EABN	Request for Express Abandonment
ELC.	Response to Election/Restriction
IFEE	Issue Fee Transmittal PTOL 85 B
IRFND	Refund Request
L RIN	Any Incoming to L&R
N417	Copy of EFS Receipt Acknowledgement
N/AP	Notice of Appeal
PA..	Change in Power of Attorney
PC/I	Power to Make Copies or to Inspect
PEF.	Pre-Exam Formalities Response
PEFRREISS	Pre-Exam Formalities Reissue Response
PEFRSEQ	Pre-Exam Formalities Sequence Reply

INCOMING

LET.	Misc Incoming Letter
IMIS	Miscellaneous Internal Document

PGEA	Req Express Aband to avoid Publication
PGA9	Req for Corrected Pat App Publication
PGREF	Req for Refund of Publication Fee Paid
PROTEST	Protest Documents Filed by 3 rd Party
PROTRANS	Translation of Provisional in Nonprovisional
REM	Applicant Remarks in Amendment
RESC	Rescind Non-Publication Request
ROCKET	Request for Design Processing
XT/	Extension of Time filed separate

APPL PARTS

371P	PCT Papers in a 371Application
A...	Amendment Including Elections
A.NE	After Final Amendment
A.PE	Preliminary Amendment
ABST	Abstract
ADS	Application Data Sheet
AF/D	Affidavit or Exhibit Received
APPENDIX	Appendix

APPL PARTS

ARTIFACT	Artifact
CLM	Claim
COMPUTER	Computer Program Listing
CRFL	CRF Transfer Request
CRFS	Computer Readable Form Statement
DIST	Terminal Disclaimer Filed
DRW	Drawings
FOR	Foreign Reference
FRPR	Foreign Priority Papers
IDS	IDS Including 1449
NPL	Non-Patent Literature
OATH	Oath or Declaration
PET.	Petition
RUSH	OPUBS Printer Query
SEQLIST	Sequence Listing
SPEC	Specification
SPEC NO	Specification Not in English

IN THE CLAIMS

Please amend the claims as follows:

Claim 1 (Currently Amended) A turbine moving blade comprising a platform having a gas path surface extending in the combustion gas flow direction, and a blade portion erecting on said platform, said gas path surface of platform being coated with a thermal barrier coating, wherein

said thermal barrier coating substantially covering said gas path surface and being is formed so as to go around from said gas path surface of platform to at least a part of the outer peripheral face of said platform.

Claim 2 (Original) The turbine moving blade according to claim 1, wherein a step portion is formed in at least a part of the peripheral edge portion of said platform, and said thermal barrier coating is formed so that it goes around to said step portion and the end face thereof is in contact with the upper face of said step portion.

Claim 3 (Currently Amended) A turbine moving blade comprising a platform, a blade portion erecting on said platform, and a shroud provided at the tip end of said blade portion, a gas path surface extending in the combustion gas flow direction of said shroud being coated with a thermal barrier coating, wherein

said thermal barrier coating substantially covering said gas path surface and being is formed so as to go around from said gas path surface of shroud to at least a part of the outer peripheral face of said shroud.

Claim 4 (Original) The turbine moving blade according to claim 3, wherein a step portion is formed in at least a part of the peripheral edge portion of said shroud, and said

thermal barrier coating is formed so that it goes around to said step portion and the end face thereof is in contact with the upper face of said step portion.

Claim 5 (Currently Amended) A turbine stationary blade comprising a pair of shrouds each having a gas path surface extending in the combustion gas flow direction, and a blade portion held between said shrouds, at least either one of said shrouds being coated with a thermal barrier coating, wherein

said thermal barrier coating substantially covering said gas path surface and being is formed so as to go around from said gas path surface of shroud to at least a part of the outer peripheral face of said shroud.

Claim 6 (Original) The turbine stationary blade according to claim 5, wherein a step portion is formed in at least a part of the peripheral edge portion of said shroud, and said thermal barrier coating is formed so that it goes around to said step portion and the end face thereof is in contact with the upper face of said step portion.

Claim 7 (Currently Amended) A turbine split ring having a gas path surface extending in the combustion gas flow direction, said gas path surface being coated with a thermal barrier coating, wherein

said thermal barrier coating substantially covering said gas path surface and being is formed so as to go around from said gas path surface to at least a part of the outer peripheral face.

Claim 8 (Original) The turbine split ring according to claim 7, wherein a step portion is formed in at least a part of the peripheral edge portion, and said thermal barrier coating is

formed so that it goes around to said step portion and the end face thereof is in contact with the upper face of said step portion.

Claim 9 (Currently Amended) A gas turbine for producing power by expanding a high-temperature and high-pressure combustion gas by using a turbine stationary blade and a turbine moving blade, wherein

 said turbine moving blade comprises a platform having a gas path surface extending in the combustion gas flow direction, a blade portion erecting on said platform, and a thermal barrier coating for covering said gas path surface of platform, and said thermal barrier coating substantially covering said gas path surface and being is formed so as to go around from said gas path surface to at least a part of the outer peripheral face of said platform.

Claim 10 (Currently Amended) A gas turbine for producing power by expanding a high-temperature and high-pressure combustion gas by using a turbine stationary blade and a turbine moving blade, wherein

 said turbine moving blade comprises a platform, a blade portion erecting on said platform, a shroud provided at the tip end of said blade portion, and a thermal barrier coating for covering a gas path surface extending in the combustion gas flow direction of said shroud, and said thermal barrier coating substantially covering said gas path surface and being is formed so as to go around from said gas path surface of shroud to at least a part of the outer peripheral face of said shroud.

Claim 11 (Currently Amended) A gas turbine for producing power by expanding a high-temperature and high-pressure combustion gas by using a turbine stationary blade and a turbine moving blade, wherein